

April 12, 1930

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AVIATION

The Oldest American Aeronautical Magazine



Rate of Growth OF THE INDUSTRY

Two Way Radio DEVELOPMENT

Why THEY *Spin* THE WAY THEY DO

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2. Distance of 500 kilometers at 160,114 miles per hour
3. Distance of 1000 kilometers at 152,702 miles per hour

WITH 500 KILOGRAMS LOAD

4. Distance of 100 kilometers at 185,62 miles per hour
5. Distance of 500 kilometers at 171,280 miles per hour
6. Distance of 1000 kilometers at 152,702 miles per hour

Recognition by the Fédération Aéronautique Internationale of these new records will credit the United States with 25 senior flight, heavier than air world's records. Pratt & Whitney will hold 17 of these, or more than half of this country's heavier than air records, and these times as many as are held by any other American aeronautical engine manufacturer.

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Illustration shows the Model 80-A Boeing Transport which will be used by the Boeing system starting in May on route of 1939 over the San Francisco-Oakland. Chicago air passenger and mail service. The Boeing system has a record of more than 9,000,000 miles of flying, and in most of these 9,000,000 miles Belden Aircraft Wire has been present.

In designing the Boeing Model 80-A Tri-Motor Transports—designed for daily service between Chicago and San Francisco—materials were selected by the most rigid specifications developed by experience acquired in 9,000,000 miles of flying.

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PLAIN FACTS from the flying text-book D. H. MOTH

MAN who fly deal in facts. Facts have little place in their profession unless they are resolvable into school performance. The following extracts from a letter written a few days ago by James H. Ritter, Operating Manager of Air Service, Inc., of Johnstown, Penn., bring to training school and commercial operators "actual performance" facts of the D.H. Gipsy Moth—facts that have made the Moth the acknowledged leader of America's training process.

"We have used the D. H. Moth over 700 hours during the last nine months without any trouble whatsoever—most of this time in dual instruction and student's solo. We have had to buy no new engine parts. At 375 hours we gave it a top overhaul and found the engine in such good shape that it would have been good for as much more time with our overhaul.

"In training, some of the students have required more than eight or nine hours of dual, although many of them could have soloed at five hours, were it not for regulations. All of our students who have completed their course—two of them women—have received their licenses. The



Moth certainly stands up under student instruction wonderfully well—better than any plane I have been my privilege to observe.

"We have loaded over 1000 paid passengers in the Moth plane—most of these because they preferred the "clotted wing" to other planes not so equipped.

"In the Moth we average about six gallons of gasoline to the hour, and counting oil changes, about one pint of oil. With that kind of performance our results on a small field 1750 ft. above sea level, it is only natural that we should be somewhat biased on the side of the D. H. Gipsy Moth over any other light ship we have."

With 102 m. p. h. speed the Moth has a light ship's easy maneuverability and smooth control—with a rugged structure under hard use which surpasses that of ships for heavier in weight and more expensive to own and operate. It is powered with the famous Gipsy engine and possesses remarkable stability through Handley-Page slatted wings and D. H. harmonized control surfaces.

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THE OLDEST AMERICAN AERONAUTICAL MAGAZINE

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EDWARD T. WARNER, Editor

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"Warning—Don't Land Here!"

UPON the editorial desk there lie a number of documents. There is an issue of the little leaflet entitled "Notions to Mariners" published every few days by the Hydrographic Office of the Navy Department. There is a recent copy of the Air Commerce Bulletin issued by the Aeronautics Branch. There is a pile of loose sheets circulated by the British Air Ministry under the collective title of "Notions to Air Men—Navigational Warnings."

The Hydrographic Office information for seamen appears as pamphlet form, but with the items so arranged that they can readily be cut apart and posted onto the light lists and coast pilots which are the coastal navigators' bibles.

The Air Ministry segregates its navigational warnings, up which every cross-country pilot should refer himself, from its miscellaneous information by printing them on pink paper instead of blue. Each notice stands out as a separate item, with its own heading upon a separate sheet, and there is no chance for overlooking it. It is the pile that we are commenting at the moment. All of these issued in the last year, there are announcements that a flying meeting is about to be held at Lympne and on a specified date and that aircraft proceeding to that field during that period should therefore use special care, that there is a temporarily dangerous area near the middle of the Ayrsway field and that it is marked with a single red flag; that Littlestone landing ground is temporarily flooded and very soft, and that special care should be used in making landings; that the flood light at Croydon will be out of action on a specified evening, owing to a change of the wiring system; and that Royal Air Force planes will be flying at night during a specified period and over a specified region, carrying

on military priorities during which no navigational lights will be shown.

The Department of Commerce does it somewhat differently. The Air Commerce Bulletin is a pamphlet of some thirty pages appearing semi-monthly. It contains reports of accidents by the personnel of the Branch miscellaneous advice for pilots, explanation of the Department's attitude towards glider operations, statistical data upon the progress of air navigation, lists of approved type certificates issued, lists of hypothetical or proposed airports, and a section entitled "Airport and Airway Changes." Within that last section there appear, with an accurate descriptive headlines and with only the faintest evidence of the division of the several items, a miscellany of information about new airports, airport changes, lighting, radio operations, and corrections to be made on charts.

We must not vote for the British system. The information needed by the pilot or navigator should be segregated from the mass of diverse material with which the Air Commerce Bulletin intrudes it. Its nature should be plainly indicated in big bold type. There should be no opportunity for the pilot whose machine does not happen to be equipped with communicating apparatus to run his eye down a list of items heavily cluttered with such terms as "telephony," exclaim "Nothing but a lot of radio bunk!" and overlook the highly pertinent announcements that the field where he is planning to land tomorrow is temporarily shut for use.

We may be in the publishing business, but we have no connection with the type-founding industry. Only the perfect altruism and a desire to see the pilot get off the safe message that is available for him moves us in our plan for more use of red ink, sixteen-point bold-face



heads, and separate inserts on colored paper, if not a completely separate and extensively cross-indexed publication, when modification and change of air navigation facilities here to be new need.

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Military Accessories

ONE of the most important of the factors contributing to the present development of the modern airplane in all its many ramifications has been the work of the specialty and accessory manufacturers. Whether in adapting a material or product previously used in other industries to an aeronautical application, or whether in the development of an entirely new technical mechanism, the specialty manufacturers have successfully responded.

This has been brought about in many cases by the assembling and developing of highly specialized engineering forces, peculiarly well versed in aeronautical requirements. The problems successfully solved in both a technical and production sense have in most instances been applicable alike to military and commercial aircraft. In some cases it has been the spur of an overestimate of anticipated commercial expansion that has led these organizations on to their present highly developed technical resources. For these specialty organizations, in the light of the recent slowing up of aircraft production, to reduce, to eliminate, or to divert to other channels these highly skilled engineering organizations would occasion an irreparable loss to aviation as a whole.

There is a field of activity inseparably linked with aviation to which their skill might well be diverted for a time to the mutual advantage of all interested activities. There is a vast class of specialized aeronautical equipment commonly referred to as military accessories. The development of such military adjuncts as aerial bombs and torpedoes, insulators, gas nozzles and gas gauges, gas meters, trigger controls, etc., are military details of major importance, whose development is often in the first instance a charge upon corps or bureaus only cooperatively associated with those directly charged with the financing of military aviation.

Progress in the field of military aeronautical accessories has often been to some extent handicapped by the division of responsibility and authority. It is believed that much might be gained by defining the fundamental military requirements and then entrusting on a larger scale than heretofore the cooperation of these skilled specialty organizations, some of whom at least temporarily have time to spare. At the very minimum, a definite gain in the applicability of details to modern production facilities and processes should be the consequence. Government laboratories and research bureaus are splendid agencies for the development of funda-

mental ideas, but they are not, by their very nature and equipment, suited to the adaptation of their fundamental discoveries to the exact modern detail production practice. Yet it is often the failure of a comparatively insignificant detail that brings about the abandonment of an otherwise promising development.

The present situation appears to provide a splendid opportunity to better coordinate industrial military development with the most advanced production possibilities. Much might be gained in efficiency by "cleansing up" production details of military accessories in this way, at the same time providing new potential sources of supply for military accessories in time of national emergencies. Finally, such activity would provide a logical method of holding intact those splendid engineering organizations during the present disturbed period in commercial aviation.

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Laboratory Tests

OF THE USEFULNESS and importance of laboratory test work in modern aeronautical development, little need be said here. Almost without exception, the progressive airplane manufacturer either conducts research and test work in its organized department, or in some of its shops on occasion demands, or submits certain construction or material to a professional laboratory. When such tests are intended to establish the suitability of a material or fabricated unit already agreed upon for a specific application is a given matter, the authorization and procedure are relatively simple. The tested product is either satisfactory, may be modified to be satisfactory, or is rejected.

When, however, the tests verge more on general applications, bordering on true research, the problem is more complex. Thus the question of comparison of results arises, and it is necessary, so to control the tests that results are directly comparable.

A point tested today may give correct results when boiled in gasoline and after a ten-day exposure test. One tested a year from today by boiling in water and exposure for 120 days, will likewise afford certain conclusions. But the two are not directly comparable, and when such tests run into a considerable number in the course of years, with the possibilities of many other variable factors in addition to those mentioned, no satisfactory scientific basis has been provided for drawing a final conclusion as to which is most suitable.

Similar difficulties occur in structural tests. One type of beam construction may be tested in a 12-foot section 8 inches deep, another in a 20-foot section 12 inches deep. No matter how painstakingly the findings are culled, the result is seldom conclusive.

Standardized procurement tests are a step in the right

direction—much can be gained by their natural extension. Their range, however, is decidedly limited—the major part of laboratory test activity falls outside their scope.

A test can be made within any single organization in as useful check of test results already available in the records, first to determine the actual desirability of the standardized test and second to compare the methods with those previously employed. One case at this war zone, changing personnel, engines, and cycles of industrial popularity often sees tests when the desired facts might have been reasonably predicted from previous experience or established by only one or two since tests instead of a complete investigation.

The definition of the procedure for laboratory tests in one of the few fields of engineering activity where originality is not at a premium. Where a test procedure has already been established for a particular class of test, it should be followed in subsequent tests and deviated from only with the best of reason. Variations in procedure may appear highly advantageous on rapid strategy, but they may also entirely eliminate all value from the particular test as a recorded base for future general comparisons.

Besides adhering to past procedure in the basic organization, reference should also be made to existing government test procedures and to private results upon which reports are publicly available. By a little forethought, and by the simple expedient of following an established test procedure, a particular material or fabricated unit may be automatically made a part of an extensive comparative research already available. Given proper economic management, a great deal of research work can be profitably performed in the laboratory, thereby not only saving unnecessary costs but also clearing the decks for really essential test activity.

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Is Flying "Learned" or "Taught"?

IT IS PRESUMABLY the object of the aircraft industry, or at least of that part of it most anxious to sell airplanes to private owners, to make flying easy and interesting. The operation of aircraft does not concern the reach of the multitude until the airplane is conceived that it does so. So long as the operation of an airplane appears to stand among special and extraordinary mental skills, not to be explained but to be acquired only by careful observation of an expert and gradual absorption of his technique, aviation cannot be popular. The giving of flying instruction is a branch of aviation, but it is also a branch of education, and it is far more the latter than the former.

There is no place in a flying school, and fortunately at least a goodly share of the well-organized schools

realize it, for the expert pilot who has nothing to offer except his experience. If you have to choose between two men, one a transport pilot with ten years' experience and with no understanding of educational methods and no inherent sympathy with the difficulty of languages, the other a private pilot with ten hours solo but a real teacher either by native instinct or by practice acquired in teaching French or arithmetic, the second man in war months' time he could be a much better flying instructor than the first.

We force with concern that at this point we mind contended is the spot of many of our readers. The requirement here already given the magazine writer with the indignant assertion "There'll be no more pilots a bunch of hoochies!" Educational theory is a forbidding phrase.

To achieve the challenge, as barely aware that there we have an idea of limiting the teaching of flying to graduates of the Harvard School of Education. Good teachers are born much more often than they are made, and we have known pilots although unfortunately not very many of them, who never had graduated from high school but who knew more about the fundamentals of education and about how to present an idea in simple and easily reasoned form than ninety per cent of the professors in any college.

There are two essentials. First, the operation of flying must be re-estimated. There must be some central philosophy. The student must see what he is trying to learn as a whole, and not as a collection of disconnected maneuvers to be performed according to an empty separate ritual. Second, there must be some rational explanation of the finding of an airplane, some connection between the theory and the practice that the student can easily grasp and remember, and some logical technique explicable in words. Most present-day pilots have learned to land a machine by receiving the vaguest possible description of the process and then going into the air with their instructor to watch what he does and try and reproduce his actions for themselves afterwards. Having played the rôle of student with a goodly number of instructors, we can wonder on the fingers of one hand those who have had any direct and comprehensible description of the finding process to offer or any standard and simple trick for uniformly accomplishing a good landing. Those who have not had wit enough to derive anything of the sort for themselves may be furnished with it by a school manual, but those who have worked it out on their own account have proclaimed themselves competent teachers.

There ought to be far more discussion of teaching methods in the aviation industry than there is at present. There ought to be some articles written upon that subject. Teaching ought not to be a side-line to anything else whatever, and the men who can really teach are far too precious an asset to aeronautical development to be squandered for any other purpose.

T. A. T.-MADDUX *Two-Way* Radio COMMUNICATION

A Review of the Troubles Encountered in the Development of the System that is Giving Real Service on the Company's Lines

By G. E. EVERETT

*Assistant General Traffic Manager
Transcontinental Air Transport, Inc.*

WITHIN two weeks following the inauguration of T. A. T. coast to coast passenger service use of the radio transmitters aboard over to such an extent that the insulation caught fire and was burned to the detriment of well found hopes. Each time the fire was extinguished without damage except to the transmitter and to the confidence of the passengers. But each time it became more serious, each time it was emphasized that the aircraft radio which a leading radio corporation produced was a dismal failure as far as the T. A. T. line was concerned.

Through tests were made of all the component parts to discover, if possible, the fundamental weakness. E. W. Proctor, T. A. T. radio engineer, and engineers of the radio company made repeated trips with the transmitters and other tests were made on the ground. A number of weaknesses were discovered and attempts were made by the radio company to remedy them, most of which were successful. Fundamentally, the error had been in failing to accurately calculate the electrical stresses which various parts must endure. Other errors lay in the method of design, construction, and installation.

Mr. Proctor compiled the various weaknesses of the transmitter somewhat as follows: (1) complexity of design; (2) multiplicity of parts; (3) inaccessibility; (4) poor insulation of parts; (5) improper calculation of electrical stresses; (6) little attention given to the needs of aircraft installation.

In each of the six cases of fire aboard the planes inspectors disclosed insulation burned to a crisp in the power amplifier and—caused, Mr. Proctor believed, by electrical currents arcing across the terminals or between the terminals and the outside metal of the case.

In a few cases, insulation of the various relays, collapsed under the heavy load carried and burned in addition to the amplifier coil. The tube sockets arced across in still other cases, the terminals failing to carry the 1,000 volts. The tubes and sockets naturally were ruined.

All in all, T. A. T. had no radio transmitter. Much



E. W. Proctor, T. A. T. radio engineer, "fix the plane" aboard a T. A. T. plane.

favorable publicity had attended the inauguration of the service and one of the features of the line emphasized was the two-way radio communication. Advertising likewise had emphasized the radio feature all in good faith, of course, but there actually was no radio transmitter.

WHEN the sixth transmitter burned without apparent ability on the part of the radio company, to remedy the condition the problem was placed squarely on the shoulders of Mr. Proctor. He had supervised the installation of the radio equipment of planes of T. A. T., sister company of T. A. T., and had assisted with numerous experiments in the Bureau of Standards.

First of all, Mr. Proctor took a detailed inventory of the parts used in the radio company's offering. He found it to consist primarily of four units as follows:

(1) Oscillator unit. (2) Power amplifier. (3) Loading coil. (4) Demodulator for power supply. Subdividing the units further he inventoried their component parts thus: Oscillator Unit. (1) Power amplifier coil. (2) Oscillator coil. (3) Four vacuum tubes (three UV21 and 1 UX210). (4) Two chokes with resistances varying from 2 to 35,000 ohms. (5) Two relays. (6) Seven condensers with capacities varying from 1 MFD to 0.000112 MFD. (7) One pickup coil and thermocouple. (8) Two grid leaks.

Power Amplifier Unit. (1) Eight resistors varying from 2 to 35,000 ohms. (2) Three multiple contact relays. (3) Two filter condensers (0.002 MFD). (4) Two modulation transformers. (5) One transformer used as filter choke. (6) Six circuit test jacks.

Loading Coil Unit. This unit was in effect a tapered variometer the stator of which was G in diameter and 14 in. long. It was housed in a box 16x11x10 in. and weighed 13 lb.

Demodulator Unit. The demodulator with its starting panel weighed slightly over 23 lb.

With the small accessories including microphones, control boxes, antenna and antenna reel, fan leads, keys, wiring and control cables the transmitter alone weighed 305 lb.—more than the weight of the average passenger! Because of its weight and bulky arrangement, it was necessary to make the installation in the forward part of the passenger cabin utilizing the space intended for one more chair. In other words carrying space for one passenger was sacrificed to the radio transmitter. In fact, the space of the chair was not sufficient to house the transmitter. In this space was only the oscillator and loading coil units, the fan lead, the antenna reel and the slack, mounting.

In the pilot's cockpit, mounted under his seat were the remote control for the antenna reel (which did not operate) two electrical remote control boxes and on the sides of the cockpit were two microphones and two keys. Parts of the receiver also were mounted in the

cockpit the remainder being in the rear of the fuselage behind the passenger cabin.

Mr. Proctor, assisted by Daniel L. Givens, assistant T. A. T. radio engineer, disassembled several transmitters and using the original parts, wherever possible, reset at their redesigning position. The scope of the activities was first at Lambert Field, St. Louis, Mo., and later at Glendale Field, California. At each place the demand was "Simplify and condense. Eliminate parts and weight."

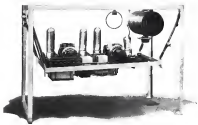
IT was day and night for several weeks. At the center Mr. Proctor had well in mind the fundamentals that must be injected into the set. The circuit to be used was the same as the commercial set had embodied, namely, the motor oscillator—power amplifier circuit. But it must be done with less than half the original parts and must be mounted on a compact panel easily accessible.

Parts were cut aside ruthlessly. It began when five relays weighing two pounds each were removed. The loading coil was cut in half. Taps and switches were removed. The two modulation transformers were removed and use of greater efficiency and less weight obtained. All boxes and cases in which the original set was housed were eliminated.

The results, finally designed and now installed on all T. A. T.-Maddux planes, is illustrated in the accompanying photograph. The circuit remains as before—motor oscillator—power amplifier and the tubes are as before—three UV21 and one UX210. Otherwise, nothing remains of the original set which our courier had to leave in orbit with the Pyrenees.

The new set has been designed to consist of only one piece; it is mounted on one removable panel hung on shock absorbers in the rear of the fuselage away from passenger interference and the passenger space formerly occupied by the transmitter has been returned to use by passengers.

The transmitter itself weighs only 29 lb. and with



The new unit radio set developed by Mr. Proctor and now used on T. A. T. planes.

power supply and all installation and operation accessories installed for remote control, it weighs but 85 lb. Pairs have been used to protect all the circuits and a five-point irreversible plug makes all the necessary connections to the transmitter including the ground. The entire set may be removed and another installed within three minutes even by a person not familiar with radio apparatus.

The three UV21B tubes serve as oscillator, modulator and power amplifier and the one UV210 serves as a speech amplifier or microphone tube. The output is 100 watts. The necessary 1,000-volt plate supply is obtained either from an engine-driven generator or a heavy-duty dynamometer. In either case a current of approximately 500 mls is drawn and distributed as follows: Modulator 100 mls, oscillator 85 mls, power amplifier 250 mls.

THE TRANSMITTER is remotely controlled from the pilot's cockpit approximately 30 ft away. A test button, however, has been installed near the transmitter to facilitate tests.

Only one sector is permanently mounted on the transmitter. This is a radio frequency meter in the antenna circuit and usually indicates a radiation of from 40 to 4.25 megacycles. There is, however, one other sector part of the radio equipment. This is a very small galvanometer on the pilot's instrument board. A two-inch loop of wire mounted near the antenna leading adjustment picks up sufficient energy to operate this meter which gives the pilot the visual assurance that everything is in order. For maintenance work and testing on the ground, elaborate plug-in test equipment is used which provides complete information relative to the condition of each circuit.

As this airplane transmitter operates on frequencies ranging between 800 and 1,300 meters a trailing wire type of antenna is necessary. A length of 375 ft is used in normal service and no resonance difficulties are ever experienced due to its fixed nature. Nevertheless, excellent contact with at least two ground stations may be had under as little as 25 ft. An extremely light weight electrically-driven antenna and a new test and maintenance

results are indicated. In the event of a forced landing, two-way contacts are always made with the T.A.T. ground stations by strapping up the antenna in the glass insulator and hand-to-hand pole carried in the tail of the plane.

BEARING IN MIND the fact that T.A.T.-Madison has approximately 7,000 miles every day in the regular transport of air passengers, it is surprising to note that only one of its two transmitters has failed during the past six months. This failure was of one of the parts of the original transmitter and need again in the redesigned transmitter. Although the airports are in the order of 425 miles apart, a plane in flight may always be sure of communicating with at least two ground stations. Absolutely continuous two-way communication is often obtained to 800 miles through considerable winds and 1,250 to 1,300 miles have been accomplished under favorable atmospheric conditions. The operator at Kingston, Ark., recently reported that a westbound pilot on the Eastern Division gave his position as 20 miles east of Columbia, Mo. Better assurance this airline director on a map.

A number of emergency uses have been discovered for the two-way radio of T.A.T.-Madison in addition to its strictly operating functions. It, of course, has been an important part of the weather bureau, pilots being warned ahead of impending weather conditions ahead. Pilots have been ordered to turn around while in flight because of operating or traffic emergencies.

As another example of two-way radio, the writer was called from bed early one Sunday morning to be informed a passenger aboard the southeast plane had suffered a request that he be met for a short interview at Lambert Field at noon. The only business was referred to the passenger who received his reply shift between Wyandak, Ohio, and Wichita and the appointment was kept.

At another time a photographic fan aboard a westbound plane contacted his supply of film between St. Louis and Kansas City. A request that additional film be waiting for him at Kansas City was met and film was supplied. This particular example is more remarkable from the fact that the pilot asked him to make the request to Wyandak and had been stranded to Kansas City by telegraph because the Kansas City station is a government station and prohibited from handling messages of that nature.

PROBLEMS were their broken during the last week market break and recovered replies showed the prices, interests, appointments have been made recently all in addition to the usual burden of necessary company communication between ground stations and planes aloft. It is perhaps interesting to note that T.A.T.-Madison radio equipment is used 90 per cent for weather information, 9 per cent for the company's business, and 1 per cent for the convenience of the passengers.

Rate of Growth OF THE AIRCRAFT INDUSTRY

By ROBERT R. DOANE

MUCH as we might want to believe certain things to be true, as they are related to the health of human endeavor in which we may happen to be engaged, yet from the standpoint of rational fact we are brought to remember that any industry, regardless of its character, when entering the economic structure soon establishes a rate of growth characteristic of itself and which is in more or less equilibrium with the ability and desire of the entire economy to receive it.

In any economic series where reliable data is available over a period of years it is a relatively easy matter to find a "trend" or rate which will give us some sort of perspective on the industry in question, as well as to serve as a basis of reasonable expectation in the future. The figures themselves, as they tabulate the final periodic results, are nothing more than the embodiment of the combined engineering, manufacturing, mechanical, financial and educational skill and force behind the progress of the enterprise. In such a method of cold calculation the elimination of bias is achieved and a rational basis for future calculation is established. When such an approach may accomplish, lies in the fact that it tells the story as truth only, without optimism or pessimism, and with a minimum of delusion for all who may be interested. As this has been the method

The rate of growth of the American aeronautical industry is a subject that has been discussed pro and con many times since Lindbergh took off for Paris. Some people argue that the rate of growth of the aircraft industry has been considerably greater than the growth of the automobile industry in the early days. Others contend that there is no basis of comparison between the old and the new transportation industries. In this article Mr. Doane treats that subject in a most interesting way and brings out many facts that should be of value to our readers.

commonly employed in measuring other industries it would seem not without reason to employ it in the aircraft industry.

IN ALL SUCH long term measurements due allowances must be made for fluctuating conditions both internal and external to the industry. In some years technological improvements creating an accelerated demand will naturally cause the volume to exceed the actual rate of growth, while in other years unfavorable factors may be expected to operate in the opposite direction. Therefore the averages, as shown in Table I, reproduced here, are based upon an average "normal" year.

As estimates for any five years are more likely to agree with actual totals than is an estimate for any single year a five year average is employed. The five year method is long enough to include changes in technological advance and further avoids the undue influence of abnormal years, also a five year period compares, usually, at least one complete business cycle. The estimates in Table I are based on an average annual rate of 40 per cent, whereas by actual comparison, as observed in the accompanying Table II, we find the average annual rate to be 34.62 per cent. By observing the war period (1915, 1917 and 1918) through moving forward the line of production, as illustrated in Chart I, and in the fitting



Showing the compact radio equipment originally installed aboard T.A.T. plane

of a free-hand curve as preferred by some statisticians of wide experience (The Problems of Business Forecasting, by Warren M. Persons, Business Cycles, The Problem and Its Setting, Wesley C. Mitchell) we obtain a reasonable similarity to the rate of growth of the automobile industry (40 per cent per annum) as established over the first sixteen years of its experience.

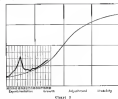
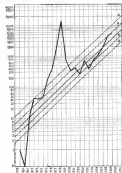
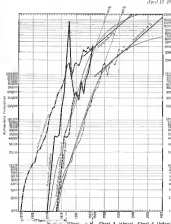
It is evident from this brief examination of the facts that long-term predictions of trends is likely to be as inaccurate as it would be to guess. But for purposes of short-term prediction (first year average period) we can compute the present normal in comparison with the normal of past years with a reasonable degree of accuracy.

In so doing for the next few years the percentage variation above or below the normal expectancy is any one's guess, with those leaders of the industry in possession of minute month-by-month figures being in an advantageous position to forecast with great accuracy the course of the immediate future. The difference in results lying in the adequacy of the statistics at hand and the way in which they are used.

CHART 2 illustrates graphically the rate of total aircraft production from 1909 to 1930. The heavy black line running diagonally across the chart in the center of the dotted line represents the annual rate of growth, and is moving forward at the rate of 34.07 per cent per year. Between the first two dotted lines marked B and B', which gives a range of 60 per cent above, and 30 per cent below normal, may be considered as the "zone of probability." Between lines A and A' a much larger percentage variation is allowed. This range may be fairly called the "zone of possibility." Under normal trade conditions the movement will be all halfhearted, fluctuate between the lines B and B'.

When view over a period of years, as in the automobile industry, the development of aviation has been very recent, and so also with the automobile industry the rates of increase are incompatible with the past experience of any other industrial sector, unless it be the very early development of the railroads. According to Dr. Carl Seydler, (Business Cycles and Business Measurements, by Carl Seydler, p. 32-41) of the Federal Reserve Bank of New York, from 1900 to 1925 automobile production increased 38 per cent per year, 47 per cent per year from 1906 to 1910, 35 per cent per year from 1910 to 1915, and 18 per cent per year from 1915 to 1920. Since 1920 the normal rate has declined to about 5 per cent per year. This represents a stupendous development of a new industry in fact the most outstanding growth with the single exception of the aircraft in chart which bids well to exceed the automobile rate.

The accompanying Table III of comparative rates of growth may be of interest as it includes the basic industries of America with their early, middle and more recent rates of increase. It will be noticed that in pre-



viously all trends the rate of growth is usually small with a 10 per cent rate exceptionally large and of recent years even 5 per cent standing above the average.

In any study of the rate of growth it is well to consider the so-called "Law of Growth" as originally set forth by Raymond H. Present ("Law of Growth in Forecasting Demand," Journal of the American Statistical Association, December, 1922, vol. 20, pp. 471-479). He suggests that all industries, whose growth depends directly or indirectly upon the ability of the

people to consume or in some way utilize their products, pass through similar phases in the course of their development. These four stages seem to be common:

1. Period of experimentation 2. Period of growth into the social and economic fabric, 3. Through the years when growth increases but at a diminishing rate, 4. Period of stability.

ON THIS BASIS, Dr. Present suggests that the trends of all industries may be represented by a single type curve—that yielded by the Gompertz equation. Every industry, he states, may have a different rate of growth, because no two industries have the same combination of influences. They will, however, trace the same type of curve, even though the rate of growth is different. If this is true, and there is ample evidence that it is, the aircraft industry is obviously past entering on its second stage (period of growth into the social and economic fabric) and as may be observed from a glance at the Law of Growth Curve (Chart 3) should continue to expand for at least the next decade at a rate which is considerably in excess of its established average.

Table II.

Year	Total	Method of Calculating Rate of Growth	Period
1909 - 1	1909 - 1	1909 - 1920	377
1910 - 8	1910 - 8	1905 - 1920	301
1911 - 1	1911 - 1	1905 - 1920	301
1912 - 41	1912 - 41	1902 - 1911	9
1913 - 41	1913 - 41	1902 - 1911	9
1914 - 46	1914 - 46	1902 - 1911	9
1915 - 170	1915 - 170	1902 - 1911	9
1916 - 40	1916 - 40	1902 - 1911	9
1917 - 129	1917 - 129	1902 - 1911	9
1918 - 40	1918 - 40	1902 - 1911	9
1919 - 10	1919 - 10	1902 - 1911	9
1920 - 10	1920 - 10	1902 - 1911	9
1921 - 10	1921 - 10	1902 - 1911	9
1922 - 10	1922 - 10	1902 - 1911	9
1923 - 10	1923 - 10	1902 - 1911	9
1924 - 10	1924 - 10	1902 - 1911	9
1925 - 10	1925 - 10	1902 - 1911	9
1926 - 10	1926 - 10	1902 - 1911	9
1927 - 10	1927 - 10	1902 - 1911	9
1928 - 10	1928 - 10	1902 - 1911	9
1929 - 10	1929 - 10	1902 - 1911	9
1930 - 10	1930 - 10	1902 - 1911	9

Table I.

Annual Aircraft Production Table
(Based on statistical forecasts for 1930-1935)

Year	1930	1931	1932	1933	1934	1935
1930	1,000	1,000	1,000	1,000	1,000	1,000
1931	1,000	1,000	1,000	1,000	1,000	1,000
1932	1,000	1,000	1,000	1,000	1,000	1,000
1933	1,000	1,000	1,000	1,000	1,000	1,000
1934	1,000	1,000	1,000	1,000	1,000	1,000
1935	1,000	1,000	1,000	1,000	1,000	1,000

1. Assuming a normal rate of growth of 34.07 per cent per year.
2. Assuming a normal rate of growth of 34.07 per cent per year.
3. Assuming a normal rate of growth of 34.07 per cent per year.
4. Assuming a normal rate of growth of 34.07 per cent per year.
5. Assuming a normal rate of growth of 34.07 per cent per year.
6. Assuming a normal rate of growth of 34.07 per cent per year.
7. Assuming a normal rate of growth of 34.07 per cent per year.
8. Assuming a normal rate of growth of 34.07 per cent per year.
9. Assuming a normal rate of growth of 34.07 per cent per year.
10. Assuming a normal rate of growth of 34.07 per cent per year.

Table III.

Industry	Normal Rate of Growth	Actual Rate of Growth	Period
1909 - 1	1909 - 1	1909 - 1920	377
1910 - 8	1910 - 8	1905 - 1920	301
1911 - 1	1911 - 1	1905 - 1920	301
1912 - 41	1912 - 41	1902 - 1911	9
1913 - 41	1913 - 41	1902 - 1911	9
1914 - 46	1914 - 46	1902 - 1911	9
1915 - 170	1915 - 170	1902 - 1911	9
1916 - 40	1916 - 40	1902 - 1911	9
1917 - 129	1917 - 129	1902 - 1911	9
1918 - 40	1918 - 40	1902 - 1911	9
1919 - 10	1919 - 10	1902 - 1911	9
1920 - 10	1920 - 10	1902 - 1911	9
1921 - 10	1921 - 10	1902 - 1911	9
1922 - 10	1922 - 10	1902 - 1911	9
1923 - 10	1923 - 10	1902 - 1911	9
1924 - 10	1924 - 10	1902 - 1911	9
1925 - 10	1925 - 10	1902 - 1911	9
1926 - 10	1926 - 10	1902 - 1911	9
1927 - 10	1927 - 10	1902 - 1911	9
1928 - 10	1928 - 10	1902 - 1911	9
1929 - 10	1929 - 10	1902 - 1911	9
1930 - 10	1930 - 10	1902 - 1911	9

1. Assuming a normal rate of growth of 34.07 per cent per year.
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4. Assuming a normal rate of growth of 34.07 per cent per year.
5. Assuming a normal rate of growth of 34.07 per cent per year.
6. Assuming a normal rate of growth of 34.07 per cent per year.
7. Assuming a normal rate of growth of 34.07 per cent per year.
8. Assuming a normal rate of growth of 34.07 per cent per year.
9. Assuming a normal rate of growth of 34.07 per cent per year.
10. Assuming a normal rate of growth of 34.07 per cent per year.

WHY THEY *Spin* THE WAY THEY DO

ANY AIRPLANE that can be made to fall off into a spin should have some means provided for recovery after it has spun a turn, regardless of the size or type of plane. If the plane is of such size and type that it is liable to be spun frequently, for student instruction, to lose altitude or any other purpose, recovery should be rapid, even after a prolonged spin. Although the spinning requirements of the Army, Navy, and Department of Commerce vary slightly, they are essentially the same, and are much more definite and stringent than the other flight characteristics of the airplane. It is a trick of fate that these requirements should be most definite about which the known means of recovery are the least definite.

In most cases the spin goes flat only after four to six turns of a normal spin. As the plane picks up speed in the spin, the inertia forces increase, producing a positive pitching moment. If this moment is greater than the negative pitching moment produced by the tail the angle of attack of the airplane increases until it reaches a region of stable autorotation or the flat spin. The difference between the normal and flat spins for an airplane may be so small that it is recognized only by a reversal of pressure on the stick when held in the rear position, or may be so great, that although recovery from a normal spin is rapid, the recovery from a flat spin is impossible, either because the controls are ineffective, or because the pilot passes out, due to dizziness and acceleration.

There are two methods of attack to the spin problem, to eliminate the flat spin, and to control it. Many factors that will need to eliminate the flat spin have no effect at all on the recovery from, or control of a spin after it

has gone flat. Again, one is prone to expect too small a margin, so that any small change in rigging, or a little overweight, may cause the plane to spin flat again. Even the same effect on an airplane in going into precession, or bringing out the next model—goes slower however, the center of gravity moves to the rear, reducing the longitudinal stability, and the spins get worse. Even though the margin may appear to be quite large, a small change in weight and center of gravity position may make a great difference in spins.

ON THE OTHER HAND, any thing that makes a flat spin more controllable, either by reducing the moment due to autorotation, or by providing a more positive righting moment to oppose the autorotation, is effective when the spin is at its worst, and variations in weight or balance have but very little effect on the recovery. Its primary balancing the elevator, the back pressure is very small. Although the Department of Commerce Airworthiness Representatives state that there should be no back pressure on the stick an airplane will be licensed if the back pressure is small and the recovery rapid.

For airplanes which will be spun often and are in the class required to be spun for the Department of Commerce, the second method is more logical. An airplane may be one of that class, however, so that it may be licensed without being spun. To consider one's design a success because it can be licensed is analogous to considering one's car a success because he, at present, is not in jail. Since, by the first method the chances of entering a normal spin (autorotation) are reduced and the airplane will go flat only after a prolonged spin with



FIG. 10. There is no rational reason why one can't pull out of a spin the airplane yet it is not uncommonly the case at times.

the stick held back after being overloaded or put into precession, this method is the better for a certain class. The main factors determining the characteristics of a spin are:

- (1) Static weight of the airplane.
- (2) Inertia forces of the horizontal component of the helix described by the center of gravity.
- (3) Inertia forces caused by the rotation of the airplane about its center of gravity.
- (4) Aerodynamic resistance of the airplane to translation.
- (5) Pitching, rolling, and yawing moments about the center of gravity due to the aerodynamic properties of the airplane rotating about the axis of the spin.

THESE FIVE FACTORS, clearly defined as they may be, are so interwoven in the actual development of a spin, that it is impossible to segregate them for separate study. The factors must balance each other—all one is lacking, the others must be reworked to make up, and they must all show a certain resistance. This resistance is not yet known, and in some cases one even wonders which way it is up.

The greatest part of the tests was made on a two place training plane, convertible land plane and single float seaplane, built for the Navy as shown in Fig. 1. In a land plane it went flat during the fourth turn, and recovered after holding full down elevator and full reverse rudder for four turns. As a seaplane, the figures remain nearly the same, but the rate of rotation was much greater. The purpose of the tests was to determine the changes necessary to eliminate the dangerous spinning characteristics. One would much rather start with a good spinning airplane, and try to find what makes it flat, than vice-versa—it is like trying to pour water through a funnel backwards.

Since the primary factor behind the spin is the weight of the airplane, it is obvious that the heavier the plane, the worse the spin. Tests were made in which the total weight was varied as well as the position of the center of gravity without changes in moments of inertia. In one case an increase in weight of 100 lb. in a landplane, all placed within a radius of 12 in. of the center of

An Account of the Spinning Characteristics of Several Airplanes Based on Flight Tests

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gravity, caused the airplane to reach its critical point in the spin, and go flat, increasing the time required for recovery from three quarters to three and one quarter turns. In a seaplane that already spun flat, the same ballast increased the time for recovery from three and a half to five turns.

The position of the center of gravity is important in reducing the tendency to go flat, but will not materially aid recovery from a flat spin. The weight of the airplane acting at the center of gravity, with the lift of the airplane produces a couple tending to rotate the airplane

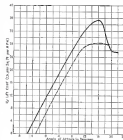


FIG. 11. The lift curve with the flat spin (dashed line) shows much better than the other.

about an axis close to the axis of pitch. The centrifugal force of the airplane acting horizontally, tending to increase the radius of the helix, and the aerodynamic reaction, produce another couple about the same axis. With a forward position of the center of gravity, the stabilizing moment is reduced, and in some cases will prevent or delay the flat spin.

Tests were first made on the effect of the center of gravity position on the airplane with the engine 15 in.



FIG. 12. This very serious spinning tendency plane proved to have a very stable spin.



FIG. 13. This plane was reported to have a very unstable spin. It is not impossible to keep it in a spin.

to the rear of the original position—thus concentrating the weight along the fuselage as much as possible. The center of gravity position was changed by the use of sweepback. It was assumed that the effect of sweepback on the spin to be compensable in its effect in normal flight producing the same results as dihedral. Tests were later made with changes in dihedral.

Tests were started with the center of gravity at 39 per cent of the mean chord, and turned forward in four stages to 26 per cent. As it was moved forward, ease came into the normal spin because more and more difficult. The airplane reached the critical point during the fourth turn, and the time required for recovery was five turns, regardless of the center of gravity position.

The most talked about cause for the spin is a precessional moment. The mass distribution is, no doubt, a large factor in a spin, but the precessional moment falls far short of being a satisfactory explanation of the flat spin. There can be no doubt, but that there is a force couple caused by the rotation of the airplane about its

and the plane was in a flat spiral. The precessional moment had virtually stalled the spin on the left side. A small commercial airplane was later tested with very nearly the same spanning characteristics. The center of gravity in this airplane was back at 39 per cent. It could not phenomena recovery, however to a staggered-dihedral cabin.

Disassembling the upper section entirely from the upper wing, and building in the wing as shown in the airplane in Fig. VI, prevented the flat spin entirely to the airplane with the long engine mount, but made no difference in the airplane with the short engine mount and with sweepback. This is not a problem in airplane design, but in mass distribution indicating that the weights along the span must be concentrated relative to the weights along the fuselage.

Reversing turns were then made, placing ballast at various points inside the section, in the tail of the fuselage, in the engine mount, in center section, and in the wing tips. Tests were all made with the same gross weight of the airplane, and the same center of gravity position. The value $I_{xx} - I_{yy} - I_{zz}$ was taken to represent the precessional moment, and $I_{xx} - I_{yy} - I_{zz}$ comparable to the weights along the span. For a given value of $I_{xx} - I_{yy} - I_{zz}$ and reducing $I_{xx} + I_{yy} - I_{zz}$ the tendency to the flat spin was reduced, and when carried far enough eliminated the flat spin, but does not reduce the time required for recovery from a flat spin. For a given value of $I_{xx} + I_{yy} - I_{zz}$ and reducing $I_{xx} - I_{yy} - I_{zz}$ the flat spin became easier, and recovery more rapid, but there was no tendency for the flat spin to be eliminated.

A thin flat airplane has a lower value of both $I_{xx} - I_{yy} - I_{zz}$ and $I_{xx} + I_{yy} - I_{zz}$ than a hullplane, and, therefore, would be expected to spin better. Reports of spin tests on conventional twin fair airplanes are not at present available, but said they are made, one can be opinionate of their results. A single thin airplane has a higher value of both, and it is well known that it will spin worse than a hullplane.

Gyroscopic forces were not considered in these tests at all. As the start in was thought airplane with right tail propellers spun worse to the left, and all test turns were, therefore, made to the left. After the tests were



Fig. 11. Structure with short engine mount and sweepback going out to test the effect of a high horizontal tail.

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would along a shock spin was made to the right, and recovery required a half turn more. Thereafter, spins were made in both directions. Skewing the spins with the propeller stalled made very little difference in either right or left spin.

Directly opposing the static weight of the airplane is the aerodynamic resistance. If the speed of descent could be reduced, the speed of rotation should also be reduced. The first tests to confirm this were made on a hullplane with a turn and a half recovery. All the interplane structural lever were removed so they were "breakable to the wind." The high speed level flight was reduced from 115 to 85 m.p.h. During about the second turn of a spin, one couldn't tell whether the plane was ever going to continue to spin or not. Even after ten turns one had to force himself to believe it to be a full spin. It recovered in three-quarters of a turn with outside held in neutral.

The same change was made on a airplane, with a turn and a half recovery. The transition from a normal to a flat spin is, at times, surprising. In the still before the spin, the wings are quiet.

As the spin picks up speed, they start to sing and clunk higher and higher in (poth) until they finally screech. When the noise suddenly drops clear down to the bottom again—and then you are flat. With the turns around as the airplane the chorus of the wings was even more correct. The noise changed very slowly till during the eighth turn, it was higher

than ever—suddenly shut—and the spin was flat as ever. In the last part of the increased resistance had been enough to prevent a flat spin. In the last part of the spin, the low favorable span distribution, the added resistance only delayed the flat spin. It was later learned that using a flat deck first and round front wings and wings, the time for recovery could be reduced from six and one-quarter to one turn.

Wind tunnel tests have been made of airplanes and wing rollers between angles of 0 and 90 deg. and the range through which autorotation is possible calculated by the strip method. Wind tunnel tests have also been made of a wing roller mounted on a spindle, and the alphas for the different types of autorotation determined.

The two tests are in fair agreement. Further tests have been made with a piece of cardboard held in a webbing net in the blast of an electric fan flat out on the ground floor. The latter was shown that the only way to keep the model from spinning, is to get the wing exactly level, and the nose exactly on the center line and even then it was possible through only a small range of angles of attack. Free flight tests show the latter quite slowly.

In a normal spin the axis of rotation is on the center line, and the wings are approximately level. If the stick is held clear back, the airplane may change as position by itself, or it may be changed by use of the elevator.

The airplane as originally designed, spins quite easily. The airplane recovered from a six turn spin in three quarter turns with neutral controls. In an airplane

to improve the flight characteristics the spin was increased 30 per cent, and a higher lift wing section used. The lift curve of the original section is the dashed line in Fig. 12, and the new section, the solid line. The loading gear was strengthened to the gross weight of the new airplane at 35 lb. less than the original.

The new model started flat in the latter part of the first turn of a spin. Recovery after one and one-half turns required one turn with several controls (full



Fig. 12. Difference between tail section of the airplane in Fig. 12 and the one shown in Fig. 13, illustrating the increase in lift curve.

down elevator and full reverse rudder). If allowed to spin longer the nose came up slowly to four turns after which recovery was possible in two and one-half turns. Longer spins were not made in that condition.

Although the increase in spin probably caused some of the trouble, it is very unlikely that it caused more than a small part of it. In a group of five single tail airplanes of almost identical dimensions and weights, the higher the maximum lift and the lower the maximum L/D of the wing section, the worse the spin.

In this case it was not desired to conduct an investigation but to apply the information gained from other tests. The stability gear was entirely successful, and the center of gravity moved forward 5 per cent. Ailerons and rudder areas were increased but for other reasons. The finished airplane is shown in Fig. 13. The spin is more difficult to start, and the recovery meets the requirements of the Department of Commerce, even though it is not in the class in which spin tests are required.

The effect of dihedral and washout was tested on a airplane in which from a flat spin, required the turns. With the center of gravity at 36 per cent, 2 deg. positive dihedral and the wing tips washed out 4 deg. by a change in rigging, the airplane could not be made to spin flat. With either the dihedral or washout removed, or the center of gravity back to 39 per cent, the plane spun flat and recovery required five turns. With this arrangement, however, the plane was so stable laterally and longitudinally, it was declared unsuitable for training purposes and was abandoned.

Changes in dihedral were made on an airplane with a short engine mount and sweepback in which recovery from a prolonged spin required three and one-quarter turns. With three degrees dihedral the spin could be started quite easily. As the dihedral was reduced, entrance into spin became more and more difficult, but with zero dihedral, it was almost impossible—the airplane spun in a half turn. After the spin was started the dihedral made no difference.

Aileron effectiveness varies with ailerons. In a good spinning aileron, the aileron profile makes but little difference in a spin. In one case in these tests as late as the aileron was held in neutral, the plane spun normally. If the aileron were set in the direction of rotation (inside stick) the plane stopped spinning, and went into a tight spin. If the aileron was held in the opposite direction, the spin went flat immediately. After the spin went flat recovery was impossible without outside stick. Invariably, the first sign that the airplane is changing its position relative to the axis of rotation, showing that the spin is going flat, is a force on the stick to the inside of the spin from the ailerons.

The First type ailerons on the upper wing with the ailerone aileron struts on the leading edge shown in Fig. 1 were entirely removed, it was impossible then to get a flat spin while recovery from a spin in the original airplane altered the spin. This was due to three things—relaxation in weight at the tips. First aileron removed, and remaining support effect of wing.

The upper ailerons on a airplane were thus removed to avoid delay in making new hinges, the old ones were used, giving an installation as shown in Fig. 5 (B). The spin would then not so flat without the use of power and outside aileron. It will be seen that the lower ailerons were inserted and the new hinges made as shown, to bring the ailerons down flush with the contour of the wing. The spins were then only slightly better than with the original ailerons, but the spin was not so flat.

With the double without modification ailerons added, the ailerons were inserted. When they were placed in their normal position the spins were flat again. In a airplane with dihedral in the tail, set with straight ailerons, recovery required one turn. Replacing the lower ailerons with a First type aileron, increased the time for recovery to one and one-quarter turns. The balanced ailerons, however, were considered to be worth the difference, and allowed to remain on the lower wings only.

A VERY RELIABLE method of improving the spinning characteristics of any airplane, is to increase the stabilizer area or area. The worst spin encountered in the tests was in the airplane shown in Fig. VI, except that the original tail shown on the airplane Fig. I was used. After only two turns, the side force on the stick had gone up very fast, and recovery was started at once by giving full reverse rudder full down elevator, and full outside aileron. All control forces were quite heavy. After eight more turns the stabilizer was pulled up to the top—first took six turns. Recovery started just before the eighth turn, and recovery was complete in ten turns. The spin was 24 turns long, and only the first two were voluntary. Yet, the attitude of the airplane had not gone beyond about 40 deg. from vertical, while the ordinary flat spin was to about 70 deg. To observers on the ground, the spin had appeared to be perfectly normal.

An increase in horizontal tail area of only fifteen per cent delayed the transition to a flat spin till after the fourth turn, and recovery was possible after a prolonged spin in five and one-half turns without changing the stabilizer adjustment. With a total horizontal tail area increase of 30 per cent, and an increase in fin area of 80 per cent, recovery could be made in four turns. On the airplane with the long engine mount, the same increase reduced the time for recovery from five to three turns. An increase in horizontal tail area is much more effective than an increase in vertical tail area.

A horizontal tail was built to a perfect Clark V section

and tested. It made no change in the spins whether right side up or upside down.

There is a theory that the fin may be blanketed in a spin by the horizontal tail. Until the angle of attack of the stabilizer has increased to the buckle point, however, the aileron follows approximately the upper surface and the fin is not blanketed. If a spin could be so flat that the stabilizer has reached with the elevator up thereby blanketing the fin, there is no use pushing the elevators down or reversing a blanketed rudder—better just sit and walk home.

Stallard was it thought possible that the reason for the ineffectiveness of the horizontal tail was that it approached, followed a wing tip, and was in the tip vortex. Since both these theories could be tested at once by raising the horizontal tail, the change was made. The stabilizer and elevators were raised a distance equal to half the wing gap. The gap left between the two halves of the stabilizer was filled with fabric. This airplane is shown in Fig. VI. The change made no difference in the spin characteristics.

The rear end of an airplane is traveling at a large angle of yaw in a spin, particularly in a flat spin. If the greatest lift from the tail is from the outside half (right stabilizer when spinning to the left) setting it at a positive dihedral angle should produce a negative pitching moment. The stabilizer was set at 6, 9, 12, 15 and 18 deg. positive dihedral on an airplane with long engine mount, and no response. With zero dihedral, recovery required three and one-quarter turns. With dihedral set at 12 deg. recovery from either direction required one and one-quarter turns. Set at 18 deg. recovery from a right spin required two and one-quarter turns, and from a left spin required two and one-quarter turns, and from a right spin required two and one-quarter turns, and from a left spin required two and one-quarter turns.

The next day a model was made. The next day the tests were repeated, and the best results were obtained with the 18 deg. dihedral. On the production airplane the dihedral is set at 10 deg., merely because it is an easier number to remember.

On an airplane with a short engine mount and steep-back that required five turns for recovery from a prolonged spin, the dihedral in the tail made no difference in spinning characteristics.

Summary and Conclusions

1. The higher the terminal velocity of an airplane the more difficult the spinning characteristics.
2. A forward position of the center of gravity is favorable at times, but is not required on all airplanes.
3. Weights along the span should be small compared to the weights along the fuselage. The better way to reduce the precessional moment is to distribute the weights more widely vertically than to concentrate them longitudinally.
4. A twin tail airplane will probably spin better than a single tail, while a single tail airplane is sure to spin worse.
5. Airplanes with high maximum lift and low maximum L/D are the worst spinners.
6. Lateral and longitudinal stability will eliminate flat spins, but may make the airplane unpleasant to fly.
7. Proper ailerons are only slightly unfavorable.
8. An increase in stabilizer area and span is the most reliable remedy.
9. Positive dihedral in what would otherwise be a horizontal tail, may produce results.

THE STORY OF



H. K. Gabel

Aircraft Tubing

The Second of a Series of Three Articles on the History and Characteristics

By S. L. GABEL,

President, Sumner Tube Co.

and HORACE C. KNERR

Consulting Mechanical Engineer, Sumner Tube Co.
President, Mechanical Laboratories, Inc.

HOT ROLL-ED tubes are the new material for the cold drawing mill. They are usually recovered by the latter in pieces about 30 ft long 2 to 4 in. in diameter, $\frac{1}{2}$ to $\frac{3}{4}$ in. wall thickness, according to the desired size of the finished product.

The first step is to cut the tube in halves or into shorter lengths. A joint in the (sometimes "long") is forged at one end, by heating a short portion of the length in a forge and hammering between dies in a stress or electric hammer. Then the tube is "pickled" in a bath of hot dilute sulphuric acid to remove scale from the inside and outside surfaces and rinsed in clear water. This pickling will often dissolve certain undesirable imperfections which have remained after hot working.

Therefore the tubes are immediately inspected for any tube defects.

The round tubes are next dipped in lime to neutralize any traces of acid, dried in an oven and lubricated with graphite just before cold drawing. Or, they may be dipped into a lubricating compound, instead of lime, and allowed to dry in air and ready for cold drawing. At the cold draw bench the point or being in passed through a die, made of very hard steel, whose internal diameter is somewhat smaller than that of the tube. At the same time a plug or mandrel of somewhat smaller

diameter than the inside diameter of the tube is inserted into the open end and pushed to a position where it will nearly coincide with the die. The point of the tube is now grasped in the jaws of a draw-head, which is caused to move forward by a powerful chain, pulling the tube through the die and over the mandrel in a continuous forward motion. This continuously reduces the inside and outside diameter and the wall thickness, and of course results in a considerable elongation of the tube.

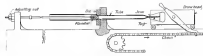
The tube has been hardened by the cold work of drawing and must go to the annealing furnace. Here it is heated under accurate control as to temperature, atmosphere and time until all the hardening effects of cold working are removed. Then it goes to the pickling



A section of the Sumner mill showing draw benches in the foreground.

back again to remove the light scale caused by annealing, is moved, cooled and lubricated as before, after which it is ready for the next pass through the cold drawing bench. This time the die and plug are again slightly smaller than the present outside and inside diameters and the tube is drawn down to a smaller size and longer length. The cycle of operations, cold drawing, annealing and pickling are repeated again and again until the final desired diameter and wall thickness have been obtained. A tube of small diameter and light wall may go through the complete cycle as many as 25 times. Of course, after a certain number of cold passes, the tube has reached a length where it can no longer be handled on the draw bench and must again be cut in half and the ends rejoined.

After reaching the final desired size, the joint or butt end and the open end of the tube, which has been water-



Diagrammatic view of draw bench showing the procedure for the process of drawing.

special shape, which cannot be passed through the rolls must also be hand straightened, sometimes with the aid of an arbor press. A machine is being designed to straighten these special shapes but many difficulties have to be overcome.

Tubes of 4130 X steel are drawn by the industry to as low as the normalized condition. These tubes therefore now pass to a normalizing furnace, where they are heated to a temperature of 1700 deg. F., withdrawn and allowed to cool with free circulation of air, giving them a tensile strength of 98,000 lb. per square inch with an elongation of 12 per cent. or more. This must be done without producing an excessive scale in the tube. When the operation is correctly controlled, the tube is cooled to a bluish black color on the surface and there is no free scale. The thickness of the oxidized surface is negligible and it acts as a mild protection against corrosion.

Special furnaces have been designed to normalize aircraft tubing. They are fired by gas and automatically controlled as to atmosphere and temperature.

Tubes of steel No. 1025 are tempered after cold drawing by heating to about 1,000 deg. F. and allowing to cool in air. This removes the extreme hardness of cold drawing and gives them a tensile strength of approx-



Graphic section of draw bench with tube in the process of drawing.

other tubes in the process are

what required, are cut off by means of a power hack saw. Special cutting wheels may be used, but these sometimes leave a thin ragged edge or burr.

The tubes are moderately bowed by the cold drawing operation, due to slight imperfections in the alignment of the die and other causes, and must be straightened. It is necessary to anneal before this straightening operation.

Several methods are available for straightening. Round tubes of relatively heavy wall may be passed through a bar straightener which rotates the tube while it passes forward between a series of diagonally placed rolls.

Another type of straightener has a series of rolls in a vertical plane staggered with each other and a similar set in a horizontal plane. The rolls are geared to fit the form of the tube, therefore this straightener may be used for square or elliptical tubes as well as round.

Tubes of very thin wall will be crushed if passed through a mechanical straightener and must therefore be straightened by hand in the old-fashioned way. The operator places the tube between stops usually mounted on a post, measures the slighting along its length and bending by hand until curves and kinks are removed. A good deal of skill is needed. Straightening tubes and others of



Tube ends in slip rolls straighter.

usually 60,000 lb. per sq. in., with excellent ductility. If normalized, this steel will develop the specified strength of 25,000 lb. per sq. in. and 22 per cent. elongation, upon which design is based, but many manufacturers find that the slightly harder tube passes better through shop



Straightening machine testing for bend with the aid of an arbor press.

operations. A final straightening after normalizing or tempering is necessary.

The tubes now are ready for the inspection department. They are laid out, a few at a time, upon an inspection table, under a good light, where they may be examined from end to end and around their entire circumference for mechanical defects, such as laps, seams, cracks, scratches and dents. By looking through the tube toward a light placed at the other end, defects on the inside may be detected.

A portion of the inspection table consists of a steel 34-inch wheel which has been placed so as to provide a straight edge 18 or 20 ft. long and 8 or 10 in. wide, upon which the tubes may be rolled to detect any departure from straightness beyond the specified limits (one part in 600). Such departure can be measured by slipping a sheet of given thickness into the space between the tube and the straight edge. Rapid and accurate inspection in this way is possible.

The tubes are checked with a micrometer as to diameter and wall thickness. Army-Navy specifications call for certain limitations as to the minimum or truly circular form of the outside of the tube, certain limitations as to the variation in wall thickness, and certain limitations as to the concentricity of the bore with respect to the outside of the tube. The wall thickness and concentricity are checked at the ends of the tube, the diameter

and accuracy of roundness at points along the length. While the tubes are being inspected for dimension and form, tensile test specimens are taken from the required number in each lot, together with such cross specimens as may be called for. Whenever possible, the tensile tests are made in full section on a piece of the tube about 10 or 12 in. long. The ends of the specimen are flanged with accurately fitting plugs having well rounded ends. This type of specimen tests the full area of the tube and gives the most satisfactory indication of its physical properties. When the tube is of too large diameter to be held by the grips, or it is beyond the capacity of the machine as to strength, a standard short metal test specimen is machined from the wall of the tube. This specimen should be machined according to Army-Navy specifications to give the best results. But when the material is well above requirements a sample specimen may be pulled or swung out a strip with parallel sides will show that specifications are met and requires less time and expense in machining. If the specimen has been taken from a tube of fairly large diameter, its conversion is not considerable and it can be gripped in the flat jaws of the machine. The slight distortion which occurs during pulling does not seriously detract from the physical properties obtained. If, however, the curvature is sharp, the specimen should be flattened before pulling. This must be done with care, using a soft hammer to avoid hardening the gauge section by cold working. If the specimen is to be bent around, this should be done after flattening, not before.

THE CRUSH TEST was devised for the purpose of detecting the presence of flaws in tubes, or a tendency to split or crush, due to increased metal. It is an excellent indication of ductility but, as it is a very severe test, it should not be applied to other than well tubing. Steel



Annealing furnace for straight tubing. The position of the tube in the heating chamber.

No. 1025 may be expected to stand the crush test in the normalized condition, but not in the tempered state, where a higher strength due to the effects of cold working, is retained. Steel No. 4130 X should be given the crush test only after annealing—not after normalizing, as the higher hardness and strength is sufficient to cause cracking in an entirely sound tube under this severe

treatment. Longitudinal flaws on either the inside or outside surface, if at all serious, will open under the crash test, and may readily be observed. If the steel is of poor quality, it will crack transversely or longitudinally in the annealed state when the crash test is applied. If surface imperfections are noted in visual inspection, regarding those annealed there is doubt, a simple cut and an indentation in the crash test will often provide the answer.

The manufacturer of high grade steel taking for aircraft steel handles two kinds of material—mild carbon steel No. 1025, and chrome molybdenum steel No. 4120X.



See mild type of workpiece for heavy steel crash tests

The 31 per cent nickel steel No. 2330, formerly used for armor, is virtually obsolete. The two types of annealing are extremely poor through production in the same mill. The greatest care is observed in the mill to avoid confusion of the two steels, but do to human fallibility, a tube of mild carbon steel may sometimes become mixed with a lot of chrome molybdenum steel or vice versa. In case the alloy steel becomes mixed with a quantity of mild carbon steel, no serious harm would result, as its properties would be superior. But if a piece of mild carbon steel became mixed with chrome molybdenum and was not detected, the danger would be serious because this steel in the normalized condition would be only a little more than as strong as normalized chrome molybdenum steel, nor would it respond to heat treatment in any important manner. Some method for marking that no mild carbon steel can go out in place of chrome molybdenum steel is therefore essential. This for a long time, constituted a very serious problem in the steel side manufacturing and also to the aircraft industry, but the problem has been solved in a satisfactory manner.

Up to the present no simple satisfactory method for rapidly separating chrome molybdenum steel tubing from mild carbon steel tubing by chemical means seems to be available. One method which has been proposed consists

in treating filings classically and observing a color reaction, but this is somewhat slow and inconsistent in a mill and is not sufficiently positive for routine inspection work. Furthermore, the results depend considerably on the physical state of the material, being indefinite when the chrome molybdenum steel has been normalized. As previously mentioned, nearly all chrome molybdenum steel tubing which is supplied to the aircraft industry is furnished in the normalized state. If mild carbon steel had become mixed with chrome molybdenum steel, it would also have passed through the normalizing treatment. The problem therefore was to find a means to distinguish normalized mild carbon steel from normalized chrome molybdenum steel. Obviously, there would be a difference in the hardness of the two steels, as well as in the tensile strength. Neither Brinell hardness tests nor scleroscope hardness tests can be used satisfactorily on thin walled steel tubing.

The Rockwell test is well adapted for the purpose, however, provided certain practical difficulties are overcome. It was found as the result of an extensive series of experiments described in an article in "True Age" of January 19, 1938 entitled "Inspection of Aircraft Tubing" that after normalizing, the Rockwell hardness of chrome molybdenum steel tubes was above 90B while that of mild carbon tubes was hardly controlled as being 80B, thus affording a sufficiently wide margin to distinguish between the two types. No direct relation between the Rockwell hardness and tensile strength of these steels has been found. The Rockwell test has been described is therefore purely a qualitative test and is as yet untried to indicate the tensile strength.

THE PRACTICAL DIFFICULTY of supporting long tubes of a wide range of diameters and wall thickness in straight alignment with the Rockwell test is great. The Rockwell test is influenced by variations of 1/16,000 in. in the position of the penetrator. Very slight misalignment or motion of the tube during the test is sufficient to spoil the reading. It is necessary to produce a clean bright surface on the inside and outside of the tube at the point to be tested, and to provide a standard extending into the open end of the tube to support its end during test. All this must be done as a precaution, so that every chrome molybdenum steel tube produced by the mill may be supported satisfactorily and without excessive cost. A machine was developed for accomplishing all of this and has been in successful operation for more than a year. Many thousands of tubes have been tested on this machine and not a single failure has come to attention, except a very few which were traceable to carelessness on the part of the operators of the machines.



Roll type straightener for square, elliptical and other special section tubes, as well as small round tubes.



The 30 lb. air driven straightening machine in which compressors and pressure straightener are attached to each component.



HIGH POINTS in the NEWS

► **Third and largest.** Annual All-American Show opens to huge new hangar at Detroit. Eighty-two planes, displayed inside with many outside in field, crash a future, first day attendance, 14,300.

► **Blue shingles forecast.** National Air Pilots Association, and manufacturers, hold meetings in Detroit concerning plans and regulations for National Air Races, Chicago, Aug. 26-Sept. 1.

► **Gliders in air.** Capt. Frank M. Hawks ends San Diego, Calif., flight 81st on Sunday afternoon, April 8, as per schedule. His record across country covered a West Coast record 2,863 mi.; gliding time, 35 hr. 47 min.

► **Many pilots trouble.** Federal authorities give out warning from \$100 to \$15,000 on all pilots, over P-W.

► **You must be notified.** Department of Commerce sets fine in minimum age for a mechanic's license.

► **A night on the Atlantic.** New York-Detroit flight in prewar 50-year Detroit is interrupted by rain from coast when night forced to land on ocean. Flight is finished next morning, however.

Airports and Airlines

► **Merge or no merge?** U. A. & T. continues fight to gain control of N.A.T., despite the latter's disavowal of merger.

► **Elton comes alive.** Starting April 26, Pan American Airways will inaugurate a seven-day mail service from New York to Buenos Aires to Montevideo in place of the biweekly schedule now maintained. Night flying will make this time six per cent shorter.

Events

► **English colonies debated.** House of Commons held arguments continuing overseas expansion, including the 1933-31 of approximately \$69,250,000, or \$4,450,000 more than last time.

Two F-32's Flying in West Coast

PETERBORO (S. J.)—Two F-32 planes, with 60 passengers for pilots, two mechanics, and two observers, left here recently for the Winn County where they will be delivered to officers. Air Express and placed in passenger service. The trip to Los Angeles is to be made with several stops.

GENERAL NEWS

HEARST'S P. DOWNS, News Editor

Sergeiev's 25,212 Ft. a Record

NEW YORK (S. J.)—Shirley test pilot Capt. Boris Sergeiev reached 24,212 ft. on March 11 in a 1935 biplane powered Sikorsky S-38 carrying a 1,000 lb. (2,204.6 lb.) load, and thereby is credited with a new world record for altitude in a biplane. This altitude has been achieved by the M.A.A., following triumphful exhibition at Rome, Germany, held the old mark of 20,303 ft., made in November 1935, in a Bristol Jupiter-powered Junkers W-34.

Minimum Mechanic's Age Fixed at Eighteen

WASHINGTON (S. J.)—The minimum age requirement for any class of Department of Commerce mechanic's license has been explained, years ago, by the Department. The minimum age requirement is now fixed at 18 years of age, as provided in Section 66 of the Air Commerce Regulations. Though the addition of the following paragraph:

"(1) The minimum age requirement for any class of mechanic's license is eighteen (18) years."

"When the Air Commerce Regulations were drawn up several years ago," Col. Young said, "the minimum age requirement for any class of mechanic's license was purposely omitted, because at that time there was no provision."

Aeronautical Finance

In this issue, AVIATION brings an important finance department which will include appear such work as the General News Section. Turn to page 774 for the serial article.

dent to grade the Department of Commerce, and those drawing up the regulations to set up a place for an airplane on the industry or the mechanics.

"Since that time experience has taught us that the minimum age requirement of eighteen years is the lowest we can permit while keeping in mind the highly important part the mechanic plays in the safe operation of aircraft and the need for mature judgment and the constant display of vigilance."

Gaelhof After Hawks' Record

WICHITA (KAN.)—Col. Arthur C. Gaelhof, son time without international record holder, will attempt, in May, to break the record which is now held by Capt. Frank Hawks. He will fly a Lockheed Vega.

All-American Show Under Way

Several Diesel-Powered Among 82 Detroit Show Craft

By Joan T. Nairn.
Special Editor of Aviation

DETROIT (WGN.)—Proceeding to be the industry's largest exposition in the diesel aircraft exposition, the third annual All-American Aircraft Show, Detroit's annual contribution to the best of aviation, opened here in a new \$1,000,000 hangar on City Airport April 8. The affair is to continue through April 13.

The show was formally opened following arrival of two Field Marshal inspectors and a Field Marshal inspector, government, and aviation officials from Field Marshal. The dedication ceremony consisted of the presentation of a huge gold key by the Charles B. Foster, Mayor of Detroit, to Col. Clarence M. Young, Assistant Secretary of Commerce for Aviation.

The key which symbolized the fact that Detroit's City Airport and hangar landside had been over given to the industry was then turned over to Edward S. Evans, chairman of the Show Committee. Edward S. Evans, City of Detroit, member of the board of directors, Detroit Aircraft Corp. and present Detroit Mayor, City of Detroit, dedicated the exposition open to the public. Among those who participated in the formal opening were Gov. Fred W. Green, of Michigan.

Many Flying on Wings

A total of 82 airplanes and 4 pilots are exhibited within the building, while approximately 80 additional planes are parked on the 250-acre airport. Inside the exhibition more 120 temporary booths, built to accommodate the 80,000 sq ft of floor space.

According to figures released by Roy Cooper, manager of the exposition, more than 14,300 persons attended the event on the first day, which, incidentally, is approximately double the attendance reported on opening day of the Detroit show last year. This would mean to indicate that the best of aviation, economy and performing airplanes at the same show in a profitable one from the point of view of average public interest.

The exhibition, itself, is considered by many to be the most important, decorated over stage. The major arrangements of the exhibition here in the exhibits, and while two guests have been given the most credit, credit that Detroit has interest in the event.

Continued on Page 744

WHAT OUR Readers Say

Non-Military Staring

To Sir Editor:
Many congratulations on your editorial in *Aviation*, of Feb. 1.
Public Non-Military Staring is one of the biggest reasons why the Aviation Industry has to face and as one who stays with aviation in this country alone can realize just how serious

Naturally most pilots like through eyes and through material necessities, to keep up the idea that a few is a superior and superior seems to be the favorite means to improve the world with this process. (And just to claim that about their escapes from death.)

I hit it, as you know, spent much of my time in France and I have attended all of the pilots on the Airfield at a thought are frequently "boiled out" for their habits and when becoming out of control or suspended change in attitude previous law and their habits the fastest of the last. The pilots are then to guide their own hand and turn in a manner to attract traffic and not to express the public with the terrible death of a pilot's life.

Another thing which does about as much damage to the exaggerated newspaper stories of crashes as the other stories and events of aviation where the pilot receives the fair maiden about the smoke of crash fire.

For many of an accident, the pilot is not, but a rough landing with fire. While I know how you feel about long landing strips, I think that you could well copy some of the speeches I have written on airport parking contracts when I first became involved with the business. The speeches which have been an absolute X, but with the experience show how on the subject of the pilot's life and the same terror." While some of the safety rules, etc., which were suggested in the old plans certainly are plus aerodynamic efficiency, their words gave a lot of surprise material to break up and about the forces of a life before anyone got badly hurt.

For while at least we are going to have some pretty rotten pilots and bad pilots to get into the air, it is not really steps and it is up to us to make it as nearly impossible for these men to kill themselves as we can.

Because of his own excellent experience my Grandfather (Paul T. & C. Lord of Civil War time) had built his life in aviation, but when I deter-

mined to enter the industrial end of the industry, such as it was in 1912, he advised me to learn to fly as I could not see the problems of design from the operator's point of view. I learned to fly a Douglas Wright and I found that the instructions of those days read like the Bible.

These things remain good to this day. I have not got on the leading edge of a wing with outstretched in front of me and taught to fly by feel, and yet by signing parts of the machine on the horizon so as to make it seem as if to tell the moment we did not have the horizon to rely on. Another thing, the instructor likes his seat used to put the compressor control over in mine and left it without power of release before we were through.

I expect for the life of me to see the second for exercise on small sport machines such as I have gotten here over my return (and which to quote our British cousin have "just the wind up"). The Klemm and Glaser which I have now stand in all over France glide with a feeling of the wind powerfully suggested in the country, it is made up of hundreds like myself. (Old air line pilots who when they get the plane started together with those who are being in-structed and who can afford up to three or four thousand dollars for a ship they own made it for a time one such year or so without sacrificing their present fortunes. However, most of us are not either rich or willing to consult with flying for a lot of the price, so called "sport" planes have.)

Many of these are excellent ships but far too tricky for our average flying ability. For most of us a good job is not to fly on a good ship, but on the only one available thing to do from a practical standpoint is to get thirty dollars, an hour or so, and fly on the best of the good side ship. Meanwhile the man who is not a pilot is not a pilot, and he has no money to spend on a good high-winged, side-by-side, non-flying, and thus, naturally, priced ship for his "sport" life.

I could say a great deal more about the industry, but for the good of the aviation industry than to induce some of the many manufacturers of light sport planes to make a few more, and spend and advertising claims for their honest state of operation, thereby creating the shell a private market source that is not very many times

one of these flights was a distance of 251 miles. This was three years before the famous flight of 1928 by M. Henri Farman and commemorated as often stated. I fall to see how the flight Farman's represented an advance in the state of the art.

It is hard enough that the French should ignore the Wright's records because they were not such champions of the F.A.I. but when they talk about history and when American engineers and American students are taught to ignore the work of the Wrights, it is very regrettable and I call upon you in the United States to do what is made appropriate now.

Georrey L. Casson
(Pilot-Engineer)
National Aeronautics Assoc.

A Light Plane Need

To Sir Editor:
In spite of your previous apology for trying to much more to say one thing, we realize that it will be many a year before *Aviation* publishes an issue half as interesting and important to the industry in general as your issue of February 1950, in which you describe the "Tanager".

One constantly reads visionary articles by aviationists who suggest explaining how to sell airplanes to a market that does not exist. There is a real potential market for airplanes, but it is not as big as the industry who can afford five to ten thousand dollar planes but are afraid to fly. To the extent possibly suggested in the country, it is made up of hundreds like myself. (Old air line pilots who when they get the plane started together with those who are being in-structed and who can afford up to three or four thousand dollars for a ship they own made it for a time one such year or so without sacrificing their present fortunes. However, most of us are not either rich or willing to consult with flying for a lot of the price, so called "sport" planes have.)

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Georrey L. Casson
(Pilot-Engineer)
National Aeronautics Assoc.

Flying Boat Facilities

To Sir Editor:
There is a point to which your attention has certainly been called before, about which I would like to put my own experience.

In my small flying boat, accompanied by an excellent pilot and co-pilot, we left Washington on Friday Jan. 30, at noon. After reaching at Hampton Roads, we landed in the evening at Norfolk City and there our experience began. We got the place at anchor but a northeastern coast during the night, the bay was very dark and the wind was so strong that I was well satisfied to let my machine sit floating even with water rolling over it.

In Charleston, where long and rain delayed us for six days, the machine was left at anchor, and in the morning it was moved every day. During that stay I went to visit some points in coastal Georgia by truck and we could not see our way before. Friday Jan. 30, at Jacksonville, where the day was worse than ever. Finally we landed in Miami the morning of Sunday Jan. 19, happy to land at last the sea and the miserable kind of the Vulture (the boat).

Starting back Tuesday noon Jan. 20, we arrived the 24th at Hampton Roads and on the following morning we found the place to Washington. Poor weather also in the return trip. It is not surprising that the point that I need to stress is the absolute lack of facilities along these over 1,000 miles of coast from Norfolk to Miami. To fly the 1,200 miles from Washington to Miami took only 12 or 15 miles, which could be made with a few small airports, and a few small airports.

In Norfolk, for instance, we had a small airport, but it was a full day's problem to get through from Washington. The weather factor is less important in a seaplane when one can fly without danger over these long stretches of

seashore water much longer than on land if you use about one quarter of a mile you can always find water in case of trouble on a fog and there are the anchor.

It is necessary that at night you should be able to find your way to the engine. It is very hard to do it in the old rain and wind in the darkness, when everything is brought to you in a white fog. I wish all of these new boats along the coast were not placed with boats. (The regular crew.) You are added to the cost of the land, as of laborers to help you to move your craft for the night to look for land.

I think that the help of all laborers on others who were to most of the time alone found by the fog never landed, and I always felt uncomfortable at their extreme willingness to help but that cannot replace the sense of a good service.

I ask your help in the name of all the seaplanes of the air (civilian or military) for fostering a campaign

aimed at the creation of bases for seaplanes. Much has been done for the land planes. A parallel case for the seaplanes should follow.

Perhaps a few on gasoline, though about gasoline is no paid at first, would help a great deal.

I enjoyed my trip immensely, but it could have been done more easily with two hours, for instance. Seaplane and Jacksonville, at Norfolk City, Charleston St. Augustine.

It may be of interest that the seaplane Flying vessel, coming down my 100 mph, and 80 mph, coming back, average 90 mph miles.

Including the use of the seaplane, Flying Boat Co. are at Miami should be multiplied. On Long Island Sound, Cape Cod and the Coast of Maine there is plenty of room for them.

[The writer is very well known to us as an expression light in the seaplane world.] He speaks that his name is withheld because of his official connections.—Ed.]

Editorial Comment FROM THE DAILY PRESS

The Aircraft's Complication

ALREADY, with the greatest of effort, to accomplish, almost as difficult to imagine as it would have been to restrict the usual strength of nations by agreement in the days when a fairly effective fighting vessel could be manufactured out of a construction by getting around the law of the sea and making her with powder and balls.

strength, an exceedingly difficult thing to accomplish, almost as difficult to imagine as it would have been to restrict the usual strength of nations by agreement in the days when a fairly effective fighting vessel could be manufactured out of a construction by getting around the law of the sea and making her with powder and balls.

—DOROTHY FINE PAPER

Metropolitan News



"Oh, yes, I'm sure the engine and all that sort of thing is all right, what I'm interested in is the airplane." —WILLIAM in the New York World



SIDE SLIPS

AVIATION
April 12, 1939

By
Robert R. Osborn

We have long realized that one of the serious shortcomings of this column was that we didn't have enough punch in it and yet we have never been able to fill this need itself being about as pointed as a Model T Ford. Also we have hesitated to broadcast an appeal for personal contributions as the majority of people have no more shilling clasp than this and we haven't done it. Now, after all these years "Good" ends in a beautiful piece, leaving the tidings of his master piece as an "We shall call it."

THESE ARE THE STORIES

In the ultimate of modern construction, all the details and gadgets have been made.

"Pilot" is the name they have chosen for the head of the school of the plane.

But consider the Keystone Pictures. Surely here's a place to meet your

I say, you can't call those things trousers. They're really two backward-pointed spots.

To be accurate in the matter, the only other picture which has appeared in this space, was written about two years ago by Mr. E.T.C. and it was really so beautiful that it is worth repeating!

THE STORY OF AVIATION

Our Henry took his chosen plane. A flying in a storm of rain. This engine trouble did develop. And now he needs a little help.

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ATTENTION TO BE USED BY THE FLIGHT CANCELLER ON ACCOUNT OF THE WEATHER.

"The Christian Science Publishing Society has now bought its second O-ring construction order on the Colonial Boston-New York service." This discovery in the Boston Herald by P.L.L.

THESE ARE THE STORIES
AVIATION IS THE

The most unusual feature about the weight of the U.S. 40 is the fact that the plane is able to carry practically its own weight when in flight.

Found in some advertising literature from the New York Show is R.R.C. of Larchmont, N.Y.

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The sight of Mr. DeW. V. B. T. of Larchmont, N.Y., spotted the following was in the annual New York Times. "A million miles of flying is scheduled for Army visitors going to and returning from the Air Corps maneuvers to be held in April at Mather Field, Sacramento, Calif., the War Department stated today."

The legendary Aviator happened to be picking up a few eggs from our desk and saw this clipping thrown. He said that a million miles were really an amount to control with, and the Department of Commerce inspectors had stated clearly that many in power to him that his journey should be provided passengers.

• • •

We have just been reading the recommendations of a U. S. Navy flight engineer to pilots who wish to indulge in high speed flying. In doing and making high speed turns he suggests that the pilot should let out a series of high pitched whistles and shrieks to aid in preventing himself from "going black." We should like to call attention to the fact that this is an old idea. Flying instructors have been telling out a series of yells and shrieks for years — with the intention of causing the student to "go black."

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Of course we have always been advising heavily in uncontrolled advertising but we do believe that it is

possible to carry honestly to the point where it becomes a loss. For instance, consider the cost sent in by L. J. M. — a clipping from the publication of one manufacturer, containing a statement: "In this new boat a series of airplanes is starting down the long assembly line, parts are coming off the line and machines and men planes will begin to appear on the airport at the end of the plant and take to their wings."

In the days of forward production a year or so ago, sometimes it was difficult to get a ship out the back door of the plant before a couple of exhaust manifolds had fallen off, but even then our manufacturers took the trouble to advertise the fact.

• • •

Mr. C.F. McR. of Los Angeles, who has lately been in the divide of the west against our article, sends us a newspaper clipping which has as its heading, "59 Army Planes Flown with Skins on 3,000 Mile Trip." He wants to know if trying a machine ship far from with skins while on a long voyage, isn't more of a stunt than has caused the last reputation of the California flyers — as they simply appear landing gears and control surfaces in flight.

We were about to tell Mr. McR. that a good business but, if there's just one more news item like this about the Army we're going to write to our reader about them. But a search for later activities which first turned us against the California flyers — it was the number of electric stoves which they were carrying in airplanes for the first time in history, which caused our editorial note.

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And if C.F. McR. wants additional proof that we have a right to wonder just what is going on in California in an uncontrolled way, let him explain the heading in the Los Angeles Times of January 12th — "Outdoor Flying Tangle."

AVIATION
April 12, 1939

THE STRENGTH OF THE PLANE IS IN THE TUBING

Reprints
of the complete series
of articles entitled

The Story of AIRCRAFT TUBING

by

S. L. GABEL, President
Summerill Tubing Company
and

BORACE C. KNEER, President
Metallurgical Laboratories, Inc.

will soon be available on request

♦

Every aircraft manufacturer and engineer will want to read and file these articles on modern aircraft construction.

We are always ready to co-operate with designers, engineers and manufacturers in solving construction problems

♦

Summerill Tubing Co.

BRIDGEPORT (Phila. Dist.) PENNA.

MANUFACTURERS OF SEAMLESS STEEL TUBING SINCE 1899



Are YOU Protected Against This Terrible Fire Danger?

KILL a fire during the first ten minutes, and the damage will be a trifling. Let it get out of control, and the loss may be staggering. And no insurance will pay for the setback to business, even more serious than material loss. The only positive protection against this fire danger is local fire fighting equipment, placed for instant use.

BADGER'S Non-Freeze 40 Gallon Fire Engine



Manual: Write for complete details and prices—TODAY!

BADGER PRODUCTS include a complete line of hand-operated fire fighting equipment in Non-Freeze, Pump, Soda-Acid, and Carbon Tetrachloride types.

BADGER FIRE EXTINGUISHER CO.
162 PARK SQUARE BUILDING, BOSTON, MASS.

AVIATION April 11, 1939



Curtiss Flying Service Hangar, Dayton, Ohio. 12
Refr. Warehouse, Shop & Refueling Gas. Aircraft
Traverse (straight and curved)

Attractive—Fireproof— Daylighted—Efficient

The specialized requirements of aeronautical buildings are fully met by standardized Truscon Steel Products at economical cost.

In the Curtiss Flying Service Hangar, shown above, Truscon Steel Doors are used throughout, including the large Seal Hangar Doors, both Straight Slide and Curved Track types.

The fireproof roof consists of Truscon Steel decks, insulated and waterproofed. The floors and roofs are supported on Truscon Steel Joists, "O-F" Open Truss and "P-G" Ply Girder Trusses. The concrete is reinforced with Truscon Welded Steel Fabric and Bars.

Whether interested in individual building products or in complete standardized buildings, write for suggestions, quotations and catalogs.

TRUSCON STEEL COMPANY YOUNGSTOWN, OHIO

See Alcoa and Alcoa in "Principles of Steel"
Truscon Steel Company, 10000, Detroit, Michigan, Ontario

Build NOW! While COSTS Are Low

TRUSCON STEEL BUILDING PRODUCTS FOR AIRPLANE HANGARS AND AIRPORTS

AVIATION April 11, 1939



Material	Specific Gravity	Modulus of Elasticity (lb./sq. in.)	Ultimate Tensile Strength (lb./sq. in.)	Yield Point (lb./sq. in.)	Elongation (in. per in.)	Impact (ft.-lb.)	Hardness (Rockwell C)	Strength (lb./sq. in.)	Strength (lb./sq. in.)	Strength (lb./sq. in.)	Strength (lb./sq. in.)
Alcoa Aluminum 7075-T6	2.8	10,000,000	70,000	35,000	15,000	30,000	11,000	11,000	11,000	11,000	11,000
Aluminum-magnesium alloy	2.8	10,000,000	70,000	35,000	15,000	30,000	11,000	11,000	11,000	11,000	11,000

Titanium Dens. Which Gives Definite Proof that Alcoa Aluminum is the One Metal that Fits Best
The chart shows strength, weight, and other properties of Alcoa Aluminum in the One Metal that Fits Best. The chart shows strength, weight, and other properties of Alcoa Aluminum in the One Metal that Fits Best.

Replacing other metals, Alcoa Aluminum offers either of these two advantages...

1. Equal strength with less weight
2. Equal weight with greater strength

Because metal will not shatter, is non-combustible, has greater strength and can be fabricated exactly and duplicated in manifold shapes, it is more and more replacing wood for plane construction.

Among the metals considered by the airplane designer and builder, one stands out. In the past few years the use of the light, strong Alloys of Alcoa Aluminum has increased tremendously, both as to the amount used in a single plane and the number of companies employing them.

These alloys have the combination of advantages so long sought for planes—both light weight and great strength. When a metal part is built

to a specified strength it can be made lighter with Alcoa Aluminum. When a part is to be built to a specified weight it can be made stronger with Alcoa Aluminum. The chart above gives technical data on this point.

The fabricating technique for Alcoa Aluminum is no more difficult than that for steel. The machinery required for either is much the same. Our nearest office will gladly supply you with full information on the application and fabrication of Alcoa Aluminum and its alloys for any definite purpose you may have in mind. ALUMINUM COMPANY OF AMERICA, 1414 Oliver Building, PITTSBURGH, PENNSYLVANIA.

ALCOA ALUMINUM



ALCOA ALUMINUM

This is one of a series of advertisements devoted originally to advertising men in an effort to make industrial advertising more profitable to buyer and seller. It is printed in these pages as an indication to readers that McGraw-Hill publishing considers men advertising effectiveness as well as editorial writing.

Every day is moving day in industry



JUST suppose we sold you the McGraw-Hill list of 600,000 paid subscribers. We wouldn't—but just suppose we did. A year from today it would be a source of complaint to you unless you also had:

- a... Our circulation staff of 100 field men who will travel nearly a million miles during 1929, checking, checking, checking.
- b... The good will of thousands of subscribers who voluntarily inform us of their changing responsibilities and addresses month after month.
- c... The Second Class postal privileges that automatically correct addresses when addressees do not.

What is a perfectly good industrial list today will be at least 25% defective a

year from today. For example: We have checked the 1928 records of 13 of the 26 McGraw-Hill publications which have a circulation of 210,176 paid subscribers.

Were it not for these channels of checking, this list of 210,176 names would have contained on December 31, 1928, deadwood to the extent of 48,120 names. Not on account of failure to renew nor delay in remitting, but because of changing functions and addresses brought about by:

<i>Plant expansion</i>	<i>Mergers</i>
<i>New projects</i>	<i>Advancement</i>
<i>Decentralization</i>	<i>Centralization</i>
<i>Out of business</i>	<i>Transfer</i>
<i>Out-and-out job changing</i>	
<i>Change of residence</i>	

McGraw-Hill PUBLICATIONS

New York Chicago Cleveland Detroit Philadelphia St. Louis
Greenville San Francisco Boston London



Where the factor of safety is all important

Through Death Valley, across the Bad Lands, the Rockies, or soaring desert sands, the reliability of Zapon products continues supreme. Since the beginning of aviation they have contributed largely in an all important factor of safety. In

Clear Mirror Aeroplane Dopes
Seven Propeller Aeroplane Dopes
Clear Painted Aeroplane Dopes

the element of safety is constantly assured by the most exacting series of continuous laboratory tests, each step vitally essential to maintaining not only the highest standard of Zapon quality but also absolute safety in the air.

Also Thinners, Lacquer Enamels and Lacquer Primers

It is the age of color runs for the plane and in keeping with this comes the Zapon. Color for interior and exterior work in all sizes, from the smallest to the largest. In color, attractive in pattern, its durability is such as to "hold and last and more."

THE ZAPON COMPANY
STAMFORD, CONN.



**Our wide
experience**
in the field of

**AUTOMOTIVE
ENGINES**

*makes us a
most practical source
for*

**AIRPLANE
SPRINGS**

*on a production or
experimental basis*

Two Plans for Spring Service

COOK SPRING CO. DIVISION
OF BARNES-GIBSON-RAYMOND INC.
ANN ARBOR, MICHIGAN
DETROIT DIVISION - 440 MILLER AVE. DETROIT



AVIATION
April 12, 1939

UP-UP-UP

The Aviation Industry's Spring Selling Drive Is Under Way

Aeronautical executives regard AVIATION as the business paper of their industry.

Each week they read it because of its up-to-the-minute news of important developments in the production, management and selling phases of their industry. The New York Show Number (dated May 2) will have even greater interest for these readers. It will picture the outstanding developments and trends evidenced at the New York Aircraft Salon in New York, May 2-6-10.

Naturally, each reader interest in this issue is bound to elsewhere, will make it an effective medium for your sales message in the advertising section.

Make Space Reservations Early

FORMS CLOSE
for complete plates
APRIL 28

AVIATION

A McGraw-Hill Publication
Tenth Ave. at 26th St. New York
Member of Audit Bureau of Circulations

AVIATION
April 12, 1939

Here's what you can learn about

AIRWHEELS



as demonstrated on a Goodyear test ship

That name "AIRWHEEL" is now well known throughout the airplane industry. It is known as the name of a tire and wheel combined—a great, yielding, tough, rolling "pillow" of rubber, built by Goodyear.

Because it is so soft (pressures range from 5 to 15 pounds) it gives a remarkable cushion to take-off or landing. It spreads out in contact with the ground, and makes landings safe even on plowed ground, on sand, on wet soggy fields—in many places where planes never dared land before.

Look at the picture and you will see that Airwheels are used for all three landing points. You will see also, that the Goodyear plane has no other shock absorbers. Think what landing gear possibilities this new product presents!

Almost every thinkable landing test

has been tried with this plane, again and again. Cross-wind, down-wind landings and take-offs. Delicate ground loops that would spell ruin with the ordinary wheel and this equipment can be performed with entire safety on Airwheels. This is due to the fact that Airwheels permit unusual side deflection and side loading as well as vertical.

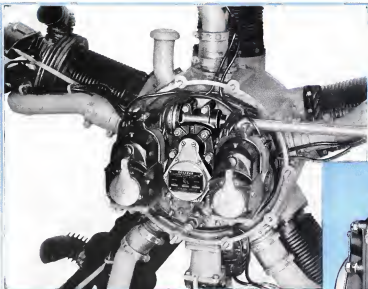
Airwheels mount directly on the hub, also made by Goodyear. Wheel flanges cannot happen, because there are no wheels. Special internal brakes, self-centering, assure the most desirable airplane landing action ever known.

Write Aeronautics Department, Goodyear, Akron, Ohio, or Los Angeles, California, for full list of sizes and for engineering data. Goodyear Airwheels are now made for planes weighing from 1000 to 20,000 pounds.



GOODYEAR

EVERYTHING IN RUBBER FOR THE AIRPLANE



KINNER K-5 ENGINE
with Eclipse Starter for quick, sure starting



Cranking Ratio 4 to 1
Weight only 12½ lbs.

A New

ECLIPSE STARTER

for the Kinner K-5 Engine

The Type SH 4 Hand Turning Gear is built in response to a universal demand for an Eclipse Starter for Kinner K-5 engine application.

The compact design of this unit, which weighs but 12½ pounds, may be noted from the above illustration in which it is shown applied to the Kinner engine. This starter's construction is such that it fits in neatly with respect to other accessories and does not add to the overall length of the engine.

The design embraces the usual Eclipse features of automatic engagement and disengagement as well as protection to both operator and starter in the event of engine backfire. The precision workmanship on this unit is reflected in its effective operation. Further particulars sent on request.

ECLIPSE AVIATION CORPORATION

East Orange, New Jersey
(Division of Bendix Aviation Corporation)